

A mixture of fuel and oxygen can be extremely



...explosive

OXYGEN



COMBUSTIBLE MEDIUM      IGNITION SOURCE

Some typical ignition sources are

- ...electrical sparks.
- ...cable fire.
- ...hot surfaces.
- ...friction sparks.
- ...static discharge.

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**FUELTANKSAFETY**

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(AM 43629)

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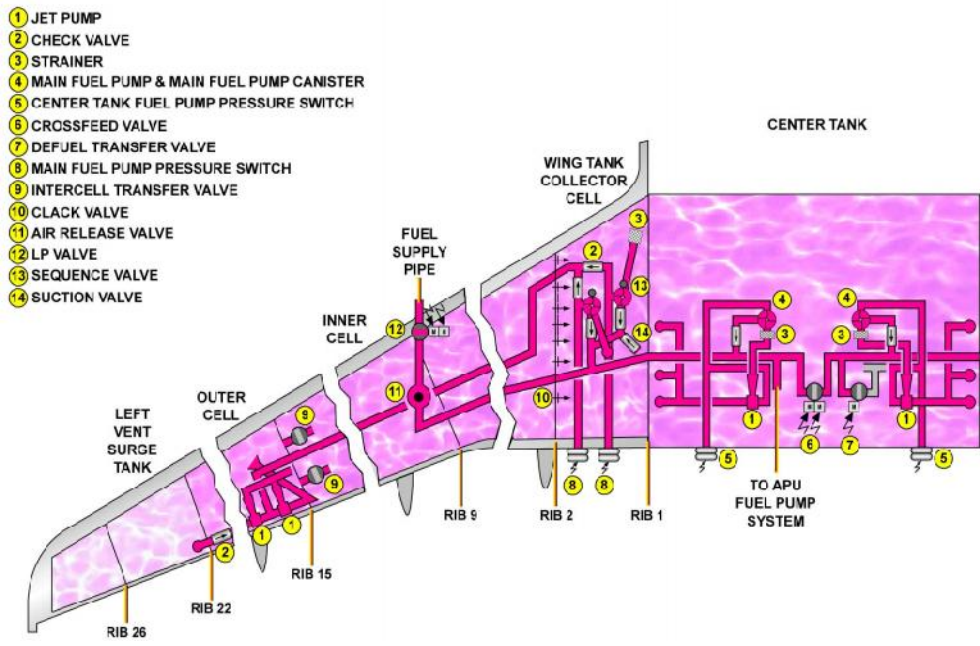
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. μ <i>TWAFlight 800 – 17/07/1996</i>	32
. μ <i>Philippine Airlines Flight 143 – 11/05/1990</i>	35
. μ <i>Thai Airways Flight 114 – 03/03/2001</i>	37
4	38
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. μ	41
. μ μ	42
5	47
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.	47
. μ <i>AIRBUS</i>	
<i>FuelTankInertingSystem – FTIS</i>	48
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2.	μ			51
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4.				52
5.				52
6.	μ	IGGS		52
7.		μ	FTIS	57
.	μ		BOEING	
			NGSSystem – NGS	57
	<b>6</b>			63
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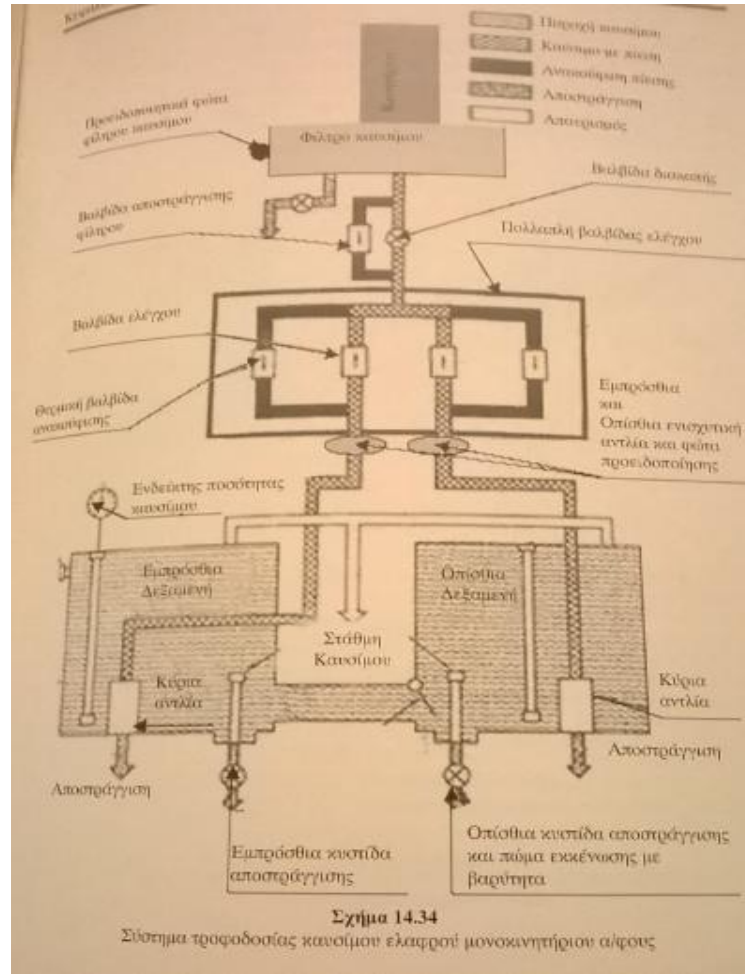
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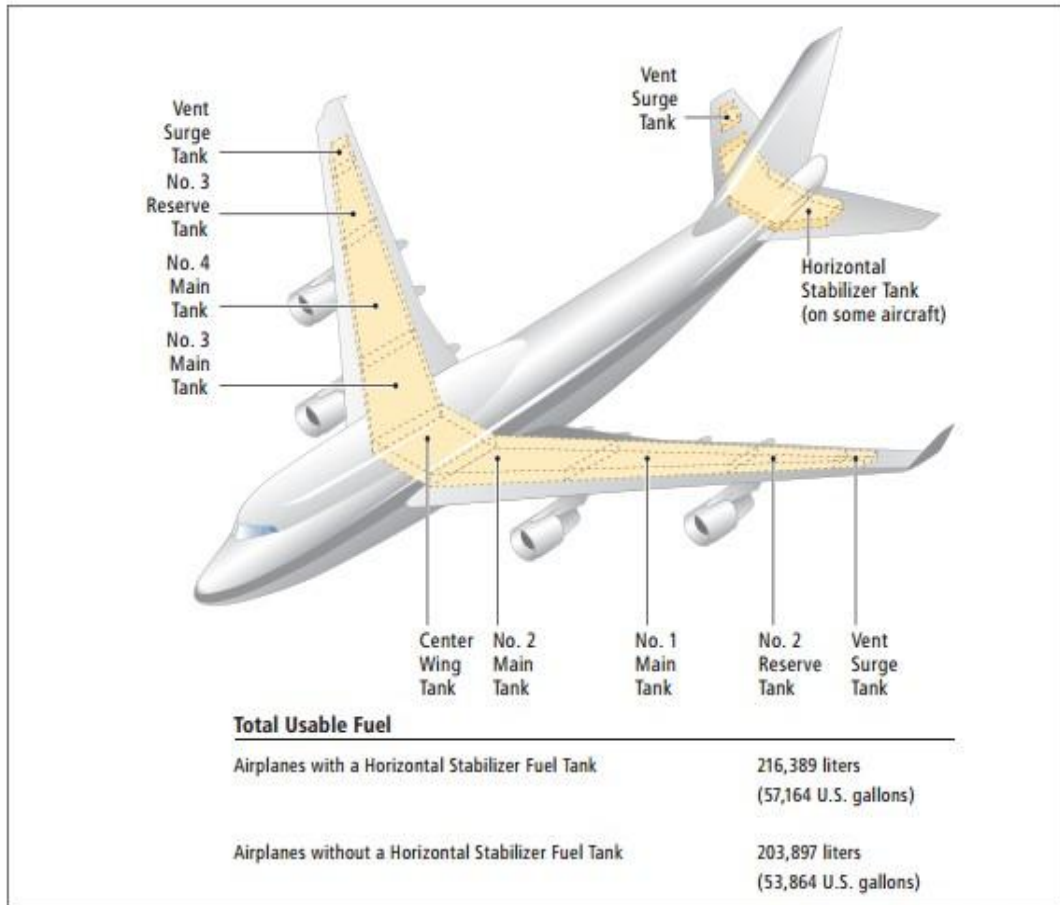
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Courtesy of Boeing

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Compound Name	Formula	Hydrocarbon Class	Boiling Point, °C (°F)		Freezing Point, °C (°F)	
n-Octane	C <sub>8</sub> H <sub>18</sub>	n-Paraffin	125.7	(258.2)	-56.8	(-70.2)
2-Methylheptane	C <sub>8</sub> H <sub>18</sub>	Isoparaffin	117.6	(243.8)	-109.0	(-164.3)
1-Methyl-1-ethylcyclopentane	C <sub>8</sub> H <sub>16</sub>	Naphthene	121.5	(250.7)	-143.8	(-226.8)
Ethylcyclohexane	C <sub>8</sub> H <sub>16</sub>	Naphthene	131.8	(269.2)	-111.3	(-168.4)
o-Xylene	C <sub>8</sub> H <sub>10</sub>	Aromatic	144.4	(292.0)	-25.2	(-13.3)
p-Xylene	C <sub>8</sub> H <sub>10</sub>	Aromatic	138.4	(281.0)	+13.3	(+55.9)
Cis-Decalin	C <sub>10</sub> H <sub>18</sub>	Naphthene	195.8	(384.5)	-43.0	(-45.4)
Tetralin	C <sub>10</sub> H <sub>12</sub>	Aromatic	207.6	(405.8)	-35.8	(-32.4)
Naphthalene	C <sub>10</sub> H <sub>8</sub>	Aromatic	217.9	(424.3)	+80.3	(+176.5)
n-Dodecane	C <sub>12</sub> H <sub>26</sub>	n-Paraffin	216.3	(421.4)	-9.6	(+14.8)
2-Methylundecane	C <sub>12</sub> H <sub>26</sub>	Isoparaffin	210.0	(410.0)	-46.8	(-52.3)
1-Ethyl-naphthalene	C <sub>12</sub> H <sub>12</sub>	Aromatic	258.3	(497.0)	-13.8	(+7.1)
n-Hexylbenzene	C <sub>12</sub> H <sub>18</sub>	Aromatic	226.1	(439.0)	-61.0	(-77.8)
n-Hexadecane	C <sub>16</sub> H <sub>34</sub>	n-Paraffin	286.9	(548.4)	+18.2	(+64.7)
2-Methylpentadecane	C <sub>16</sub> H <sub>34</sub>	Isoparaffin	281.6	(538.9)	-7.0	(+19.4)
n-Decylbenzene	C <sub>16</sub> H <sub>26</sub>	Aromatic	297.9	(568.2)	-14.4	(+6.1)

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: ChevronFuelCompanyU.S.A

Additive Type	Jet A ASTM D 1655	Jet A-1 DEF STAN 91-91	JP-4 MIL-DTL 5624	JP-5 MIL-DTL 5624	JP-8 MIL-DTL 83133
Antioxidant	Allowed	Required**	Required**	Required**	Required**
Metal deactivator	Allowed	Allowed	Agreement	Agreement	Agreement
Electrical conductivity/ static dissipator	Allowed	Required	Required	Agreement	Required
Corrosion inhibitor/ lubricity improver	Agreement	Allowed	Required	Required	Required
Fuel system icing inhibitor	Agreement	Agreement	Required	Required	Required
Biocide	Agreement	Agreement	Not allowed	Not allowed	Not allowed
Thermal stability	Not allowed	Not allowed	Not allowed	Not allowed	Agreement †

: 1.2 μ

(Additives) –

: ChevronFuelCompanyU.S.A





	Jet A-1	Jet A
Flash point	38 °C (100 °F)	
Autoignition temperature	210 °C (410 °F) <sup>[6]</sup>	
Freezing point	-47 °C (-53 °F)	-40 °C (-40 °F)
Max adiabatic burn temperature	2,500 K (2,230 °C) (4,040 °F) Open Air Burn temperature: 1,030 °C (1,890 °F) <sup>[9][10][11]</sup>	
Density at 15 °C (59 °F)	0.804 kg/l (6.71 lb/US gal)	0.820 kg/l (6.84 lb/US gal)
Specific energy	42.80 MJ/kg	43.02 MJ/kg
Energy density	34.7 MJ/l	35.3 MJ/l

:1.4 μ μ JetA-1 JetA – : Wikipedia



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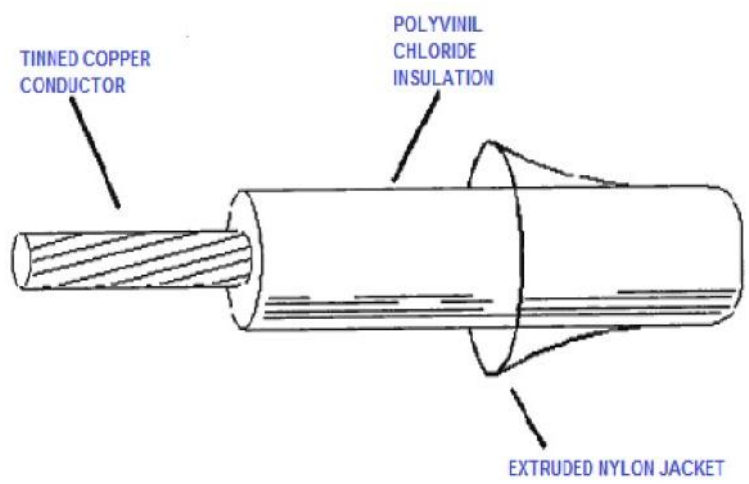
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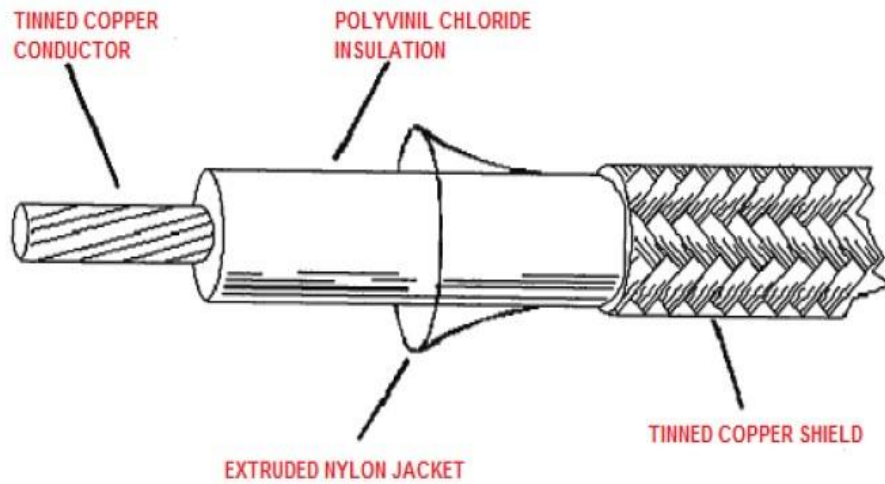
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μ :2.1 – : HATA, Materials&HardwareHandbook

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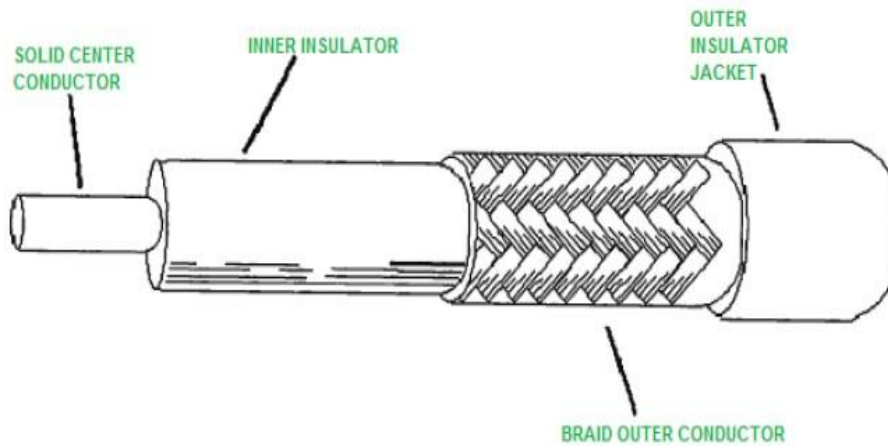
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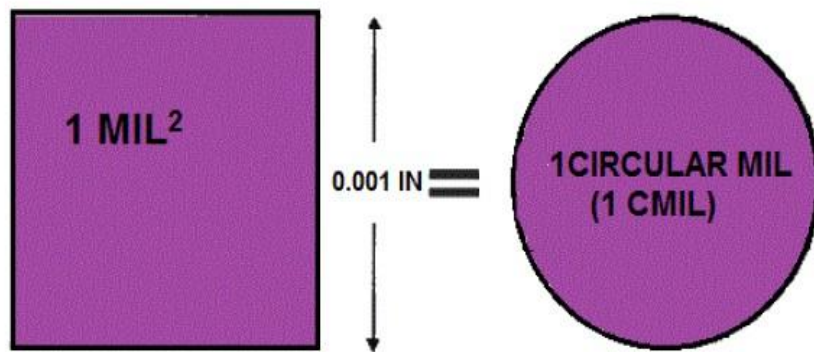
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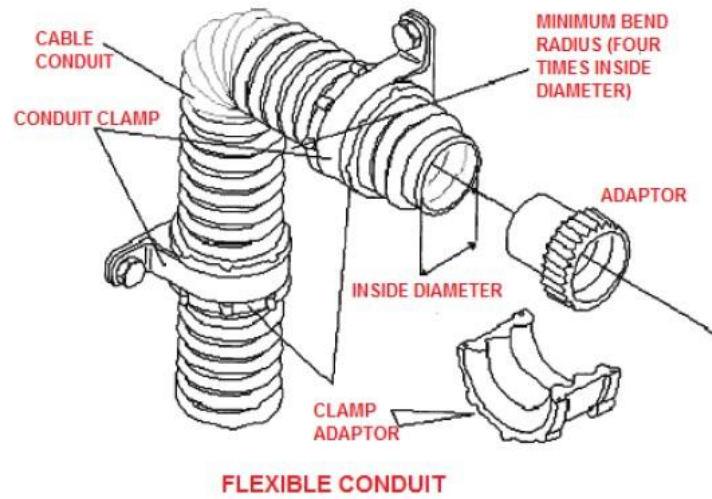
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Circular Mil – : HATA, Materials & Hardware Handbook

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- : HATA, Materials & Hardware Handbook

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	Test Condition	Maximum Resistance
Primary Conductors	Between extremities of the fixed portions of aircraft of non-metallic or composite construction.	Estimated and declared by manufacturer.
	Between extremities of the fixed portions of metallic aircraft.	0.05 ohm
	Between bonded components and portions of main earth system to which they are connected.	
Secondary Conductors	Between metallic parts normally in contact with flammable fluids and main earth system and between the parts themselves.	1 ohm
	Between isolated conducting parts, which may be subjected to appreciable electrostatic charging and the main earth system.	0.5 megohm or 100,000 ohms per sq ft of surface area whichever is the less
	Between equipment supplied from an unearthen system, of any voltage, and the main earth system.	1 ohm
	Between equipment containing circuits carrying 50 volts (RMS or DC) or more, and the main earth system.	

:2.1 – : HATA, Maintenance Practices in Aviation Handbook

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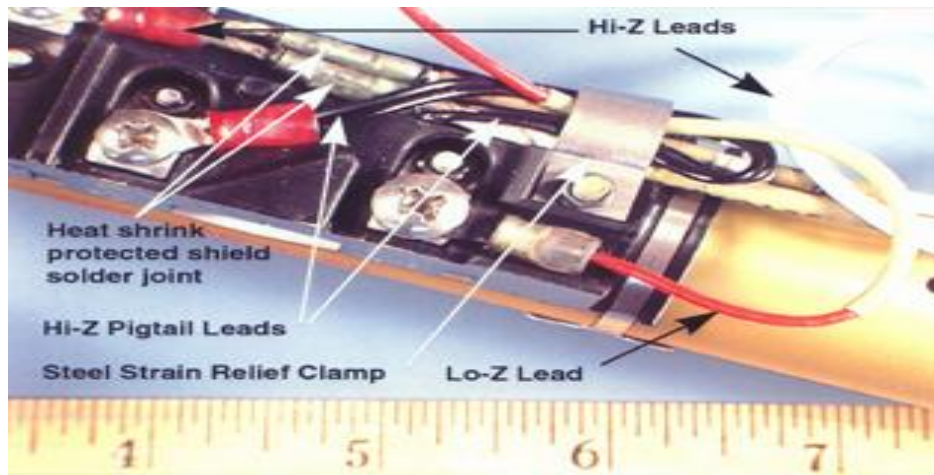
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μ **Philippine Airlines Flight 143 – 11/05/1990**

, **11** **1990** μ μ μ ,

**Boeing 737-300**

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• **μ Thai Airways Flight 114 – 03/03/2001**

3 2001, Boeing737-400 ThaiAirwaysInternational  
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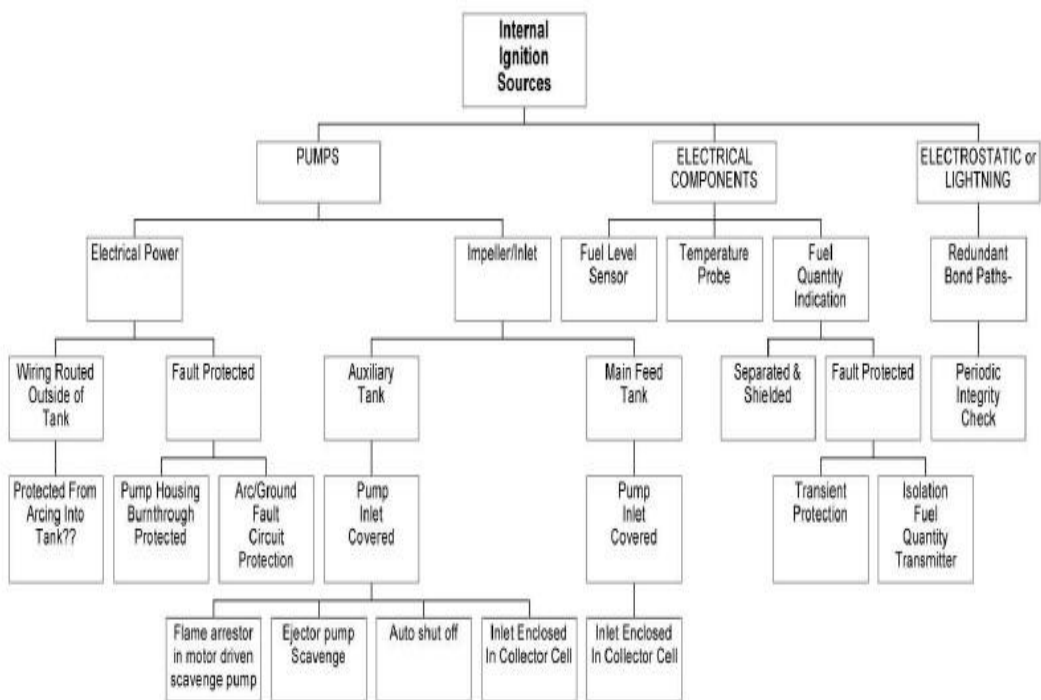
μ : 3.4 Boeing 737-4D7 HS-TDC Thai Airways - : Internet

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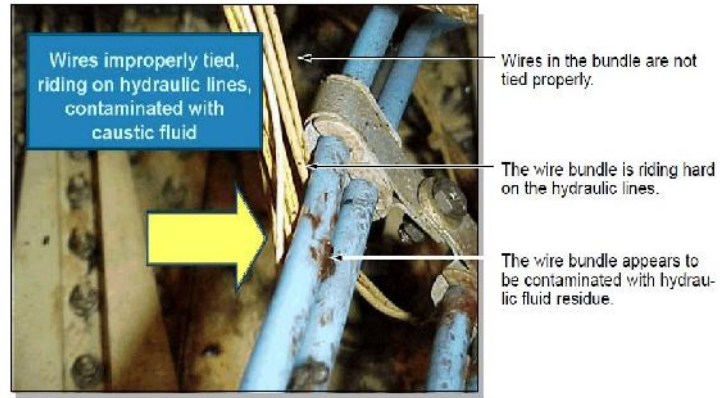
Fuel Tank Ignition Source Consideration



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 – : Lufthansa Technical Training







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– : Lufthansa Technical Training

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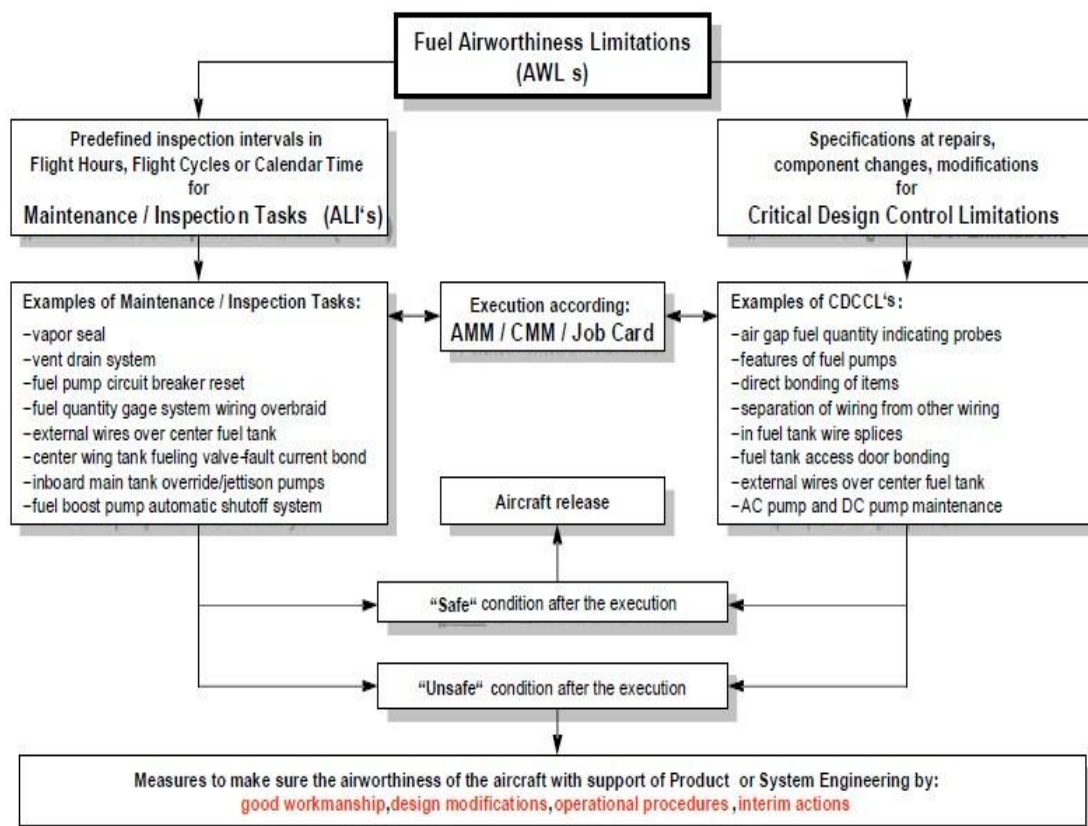
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- FAL (FuelAirworthinessLimitations):  $\mu$  (  $\mu$  ,  $\mu$  ) ,  $\mu$   $\mu$  / ,  $\mu$  « $\mu$  » .  $\mu$   $\mu$   $\mu$  ,  $\mu$  ( ) .  $\mu$

- « μ μ ».
- CDCCL (Critical Design Configuration Control Limitations):  
FAL. μ ( μ , μ , μ ), μ , μ . μ μ , μ / μ μ μ .
  - ALIs (Airworthiness Limitation Inspections): x μ μ μ ( μ μ ) μ CDCCL, μ μ , μ μ / μ μ μ .



μ : 4.4 « μ μ » (Airworthiness Limitations) – : Lufthansa Technical Training

(Maintenance Review Board Report)

(Maintenance Programs),

**Fuel Airworthiness Limitations (FALs)**

**Maintenance / Inspection tasks (ALLs)**

**Critical Design Configuration Control Limitations (CDCCL)**

**AIRBUS**  
Customer Services Directorate  
A340 Fuel Airworthiness Limitations  
**SECTION 1**  
**MAINTENANCE / INSPECTION TASKS**

TASK REF	TASK TITLE	INTERVAL	APPLICABILITY
1 26-16-00-02-1	Delayed inspection of vapour seal / drip shield	6 Years	A340-200/-300 POST MOD 40620  A340-500/-600

**AIRBUS**  
Customer Services Directorate  
A340 Fuel Airworthiness Limitations  
**SECTION 2**  
**CRITICAL DESIGN CONFIGURATION CONTROL LIMITATIONS**

CDCCL	Requirement	Reference	Applicability
1	Air gap between a fuel quantity indicating probes and the aircraft structure  Critical to prevent spark generation during lightning strike conditions and must not be compromised  The gap stated in the relevant maintenance procedure in the AMM must be achieved during installation of a fuel quantity probe.	AMM 26-42-16-400 26-42-16-400 26-42-17-400 26-42-18-400 26-42-19-400 26-42-21-400	ALL
2	Separation of fuel quantity and level indicating system wiring from other wiring.  To prevent cross coupling of high voltage transients and thereby keep unsafe ignition energies from inside the tank it is required that operators obey the wire routing and separation standards given in the Electrical Standard Practices Manual (ESPM).	ESPM 20-19-00 20-33-11 20-33-20 20-33-21 20-33-22	ALL

μ : 4.5 « μ μ μ » (Airworthiness Limitations) /  
Airbus – : Lufthansa Technical Training

**AIRBUS**  
Customer Services Directorate  
A340 Fuel Airworthiness Limitations  
**SECTION 2**  
**CRITICAL DESIGN CONFIGURATION CONTROL LIMITATIONS**

CDCL	Requirement	Reference	Applicability
1	Air gap between fuel quantity indicating probes and the aircraft structure.	AMM 28-42-10-400 28-42-10-400 28-42-17-200 28-42-10-400 28-42-10-400 28-42-21-400	ALL
2	Separation of fuelquantity and level indicating system wiring from other wiring.	To prevent cross coupling of high voltage transients and thereby keep unsafe ignition energies from inside the tank it is required that operations obey the wire routing and installation standards given in the Electrical Standard Practices Manual (ESPM) ESPM 28-19-20 28-33-11 28-33-20 28-33-21 28-33-22	

**A340**  
AIRCRAFT MAINTENANCE MANUAL  
TASK 28-42-10-400-801

Installation of the RFI, Center Tank Probe

**WARNING :** THIS PROCEDURE USES A FUEL SYSTEM ITEM THAT IS IN A CATEGORY KNOWN AS A CRITICAL DESIGN CONFIGURATION CONTROL LIMITATION (CDCL). CDCL IDENTIFIES AN ITEM THAT CAN BE THE SOURCE OF A POSSIBLE FUEL TANK IGNITION. YOU MUST KEEP ALL CDCL ITEMS IN THE APPROVED CONFIGURATION. DAMAGE, WEAR OR CHANGES TO A CDCL ITEM CAN CAUSE A POSSIBLE FUEL TANK EXPLOSION.

**WARNING :** MAKE SURE THAT THE SAFETY DEVICES AND THE WARNING NOTICES ARE IN POSITION BEFORE YOU START A TASK ON OR NEAR:

- THE FLIGHT CONTROLS
- THE FLIGHT CONTROL SURFACES
- THE LANDING GEAR AND THE RELATED DOORS
- COMPONENTS THAT MOVE.

MOVEMENT OF COMPONENTS CAN KILL OR INJURE PERSONS.

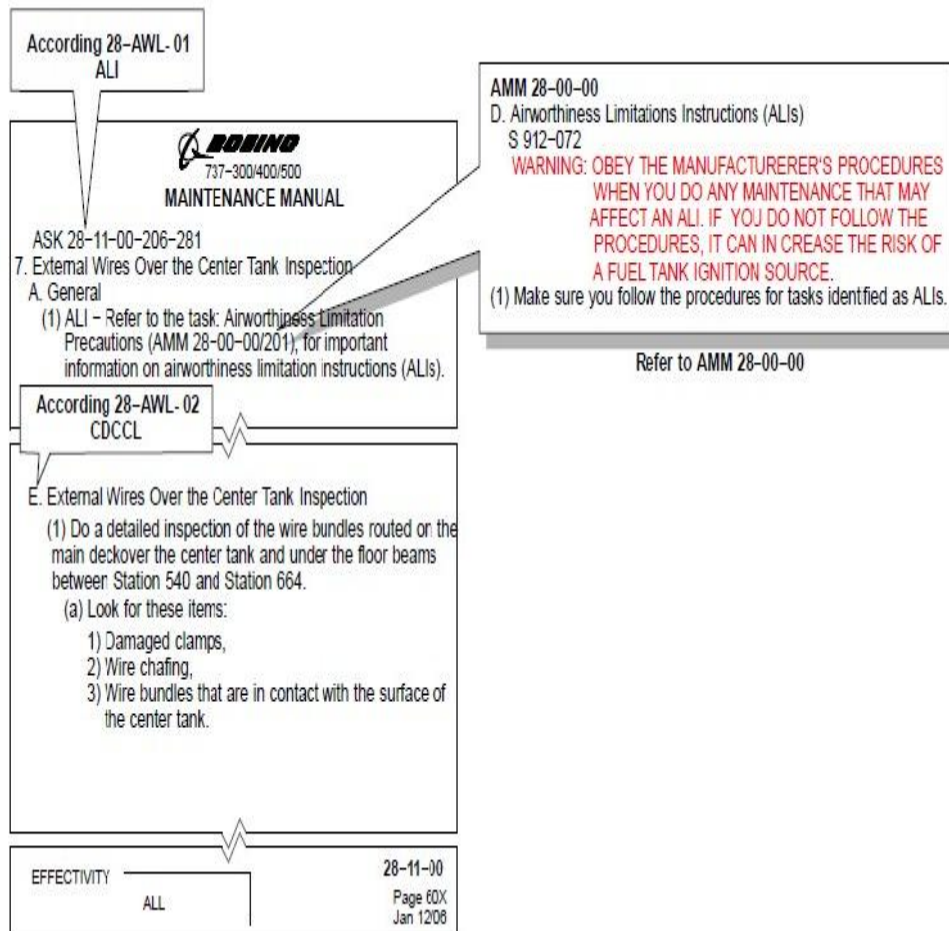
**WARNING :** OBEY THE FUEL SAFETY PROCEDURES. THIS CAN PREVENT INJURY AND DAMAGE.

μ : 4.6 « μ μ » (AirworthinessLimitations) / Airbus – : Lufthansa Technical Training

**BOEING** 737-12345 AIRWORTHINESS LIMITATIONS/CERTIFICATION MAINTENANCE REQUIREMENTS  
737-300, -400, -500 MAINTENANCE PLANNING DATA – AIRWORTHINESS LIMITATIONS – FUEL SYSTEMS

AWL NUMBER	TASK	INTERVAL	APPLICABILITY	DESCRIPTION
28-AWL-01	ALI	10 YRS/ 36000 FC NOTE	737-300/-400/-500 airplanes that have incorporated Service Bulletin 737-28-1208.	External Wires Over Center Fuel Tank Concern: Potential for Wire Chafing and Arcing to Center Fuel Tank Upper Panel. Perform a detailed inspection of the wire bundles routed on main deck over the center fuel tank and under the main deck floor boards between Sta. 540 and Sta. 663.75 to detect damaged clamps, wire chafing, damage to the vapor barrier, and that the wire bundle is not in contact with surface of the center fuel tank per AMM 28-11-00. If discrepancies are found, repair per the Boeing Standard Wiring Practices Manual (SWPM) D6-54446.
			<b>Reference: AMM 28-11-00</b>	Note: Threshold for initial inspection is 10 years after incorporation of Service Bulletin 737-28-1208. Repeat Interval is 10 years or 36000 FC, whichever comes first.
28-AWL-02	CDCL	N/A	737-300/-400/-500 airplanes that have incorporated Service Bulletin 737-28-1208.	External Wires Over Center Fuel Tank Concern: Potential for Wire Chafing and Arcing to Center Fuel Tank Upper Panel If any maintenance is performed in the area under the main deck floor boards and over the center fuel tank, verify the following: 1. Maintain the existing wire bundle routing and clamping. 2. Installation of any new wire bundles must be per Boeing Standard Wiring Practices Manual (SWPM) D6-54446. 3. All wire bundles over the center fuel tank are inspected per 28-AWL-01.

μ : 4.7 « μ μ » (AirworthinessLimitations) / Boeing – : Lufthansa Technical Training



μ : 4.8 / μ Boeing – μ » (Airworthiness Limitations) : Lufthansa Technical Training

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(OxygenEnrichedAir - ) .

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Boeing μ μ

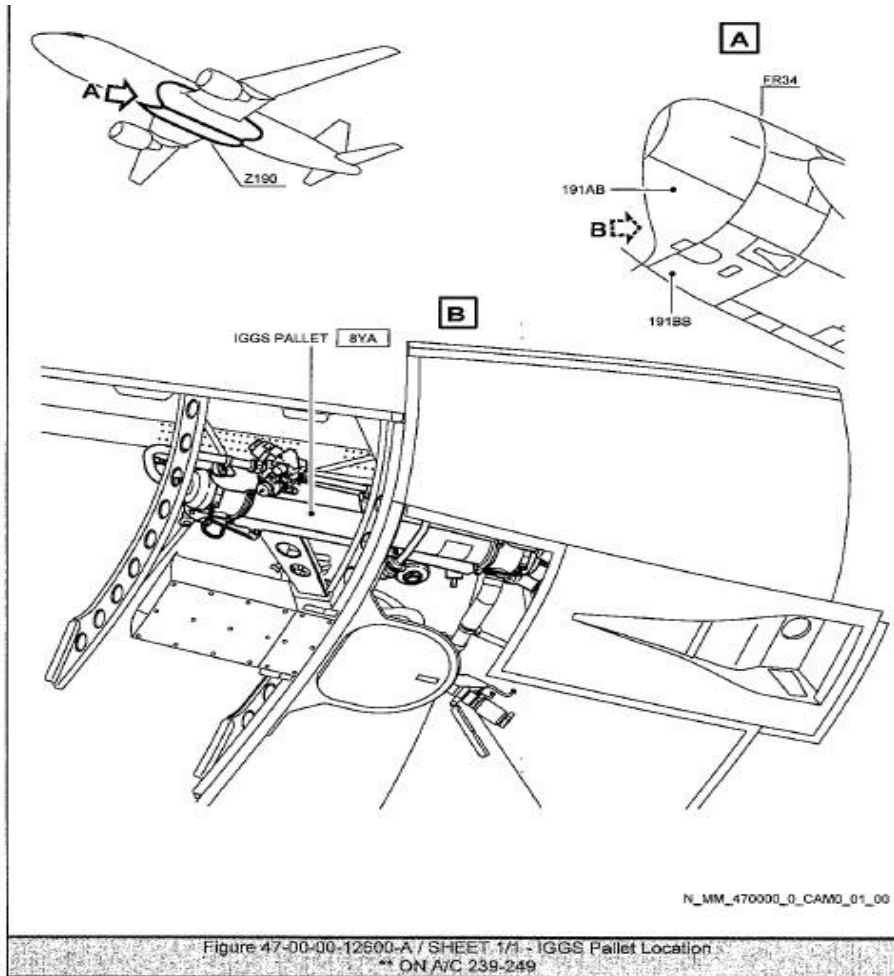
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Airbus μ

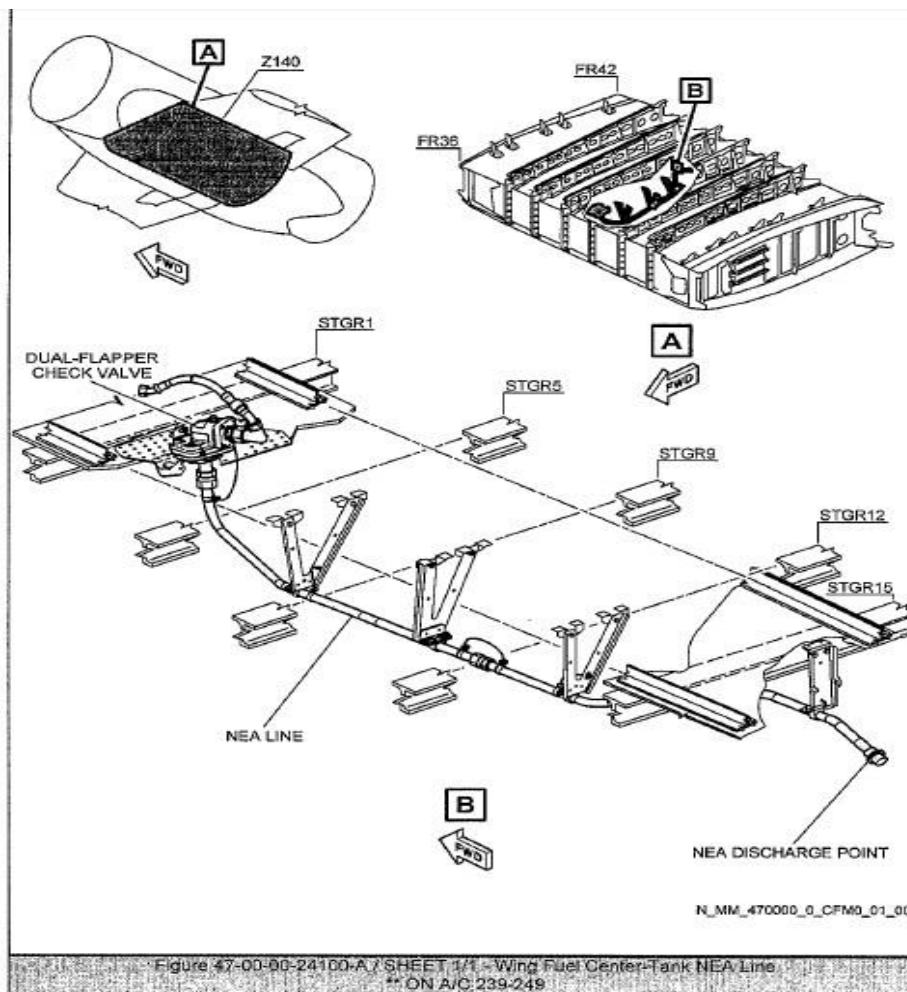




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μ : 5.1 μ IGGS – : Airbus Aircraft Maintenance Manual



μ : 5.2 - : Airbus Aircraft Maintenance Manual

1.

CSAS, IGGS. IGGS. ( «Double-UltraLowParticleAir» , (AirSeperationModule), (DualFlowShut-OffValve),

Double-UltraLowParticleAir (D-ULPA). D-ULPA (AirSeperationModule).

AirSeperationModule ( ) μ μ μ μ ( )

μ μ AirSeperationModule  
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 DualFlowShut-OffValve IGSpallet μ μ .  
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 ULPA AirSeperationModule, μ  
 μ μ AirSeperationModule  
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2. μ IGGS  
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 85 C.

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 dualflowshut-offvalve. CCU μ μ IGGS  
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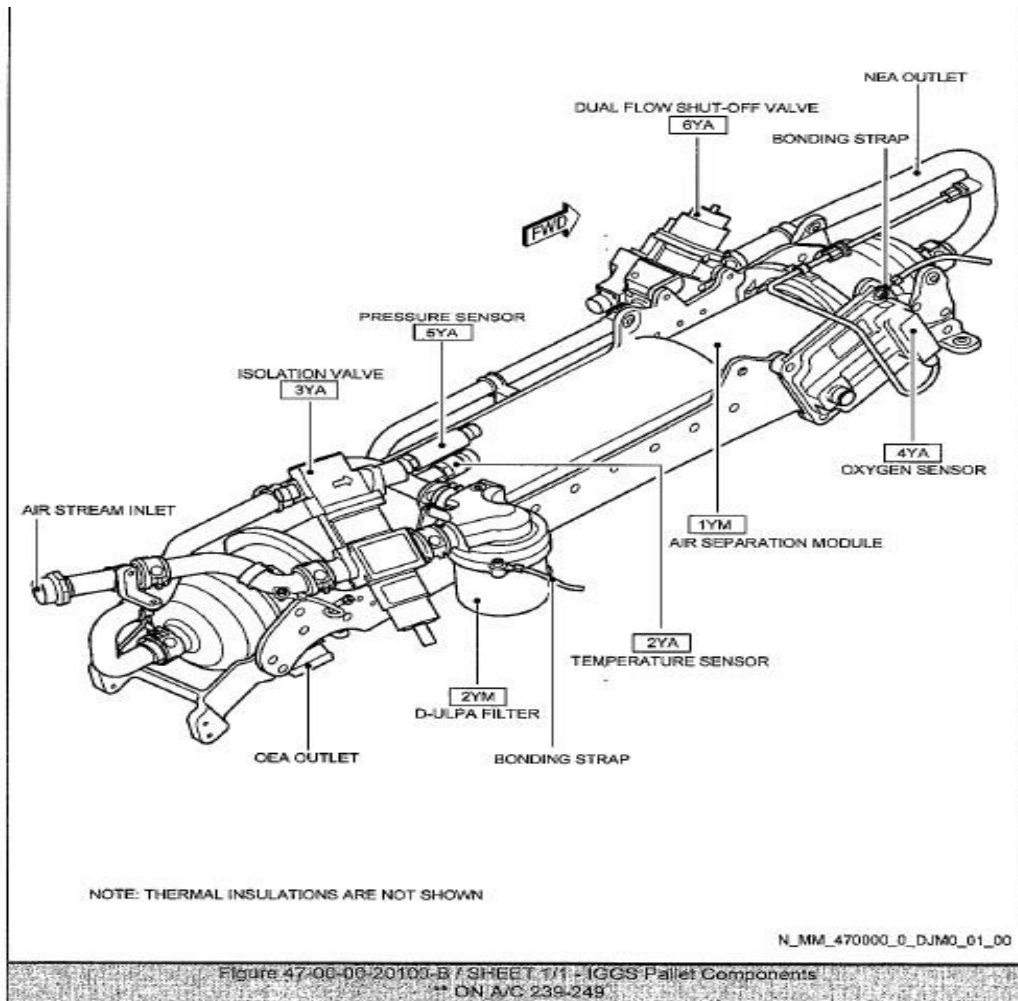
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 μ μ AirSeperationModule,  
 μ μ IGGS. μ  
 AirSeperationModule FTIS  
 (cruiseflight).

5.

IGGS. μ 103PP μ 28 VDC μ  
 μ μ IGGS 28VDC  
 μ , μ ,  
 DualFlowShut-OffValve.

6.

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 T μ IGGS μ μ Air-Condition / , μ  
 μ (DC), μ μ ADIRU.



μ : 5.3 μ μ IGGS – : Airbus Aircraft Maintenance Manual

### - D-ULPA

D-ULPA μ μ μ μ  
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 AirSeparationModule. μ μ  
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 AirSeparationModule μ 54 C.

- **Air Seperation Module**

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 μ μ μ μ μ AirSeperationModule  
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- **IGGS Pallet**

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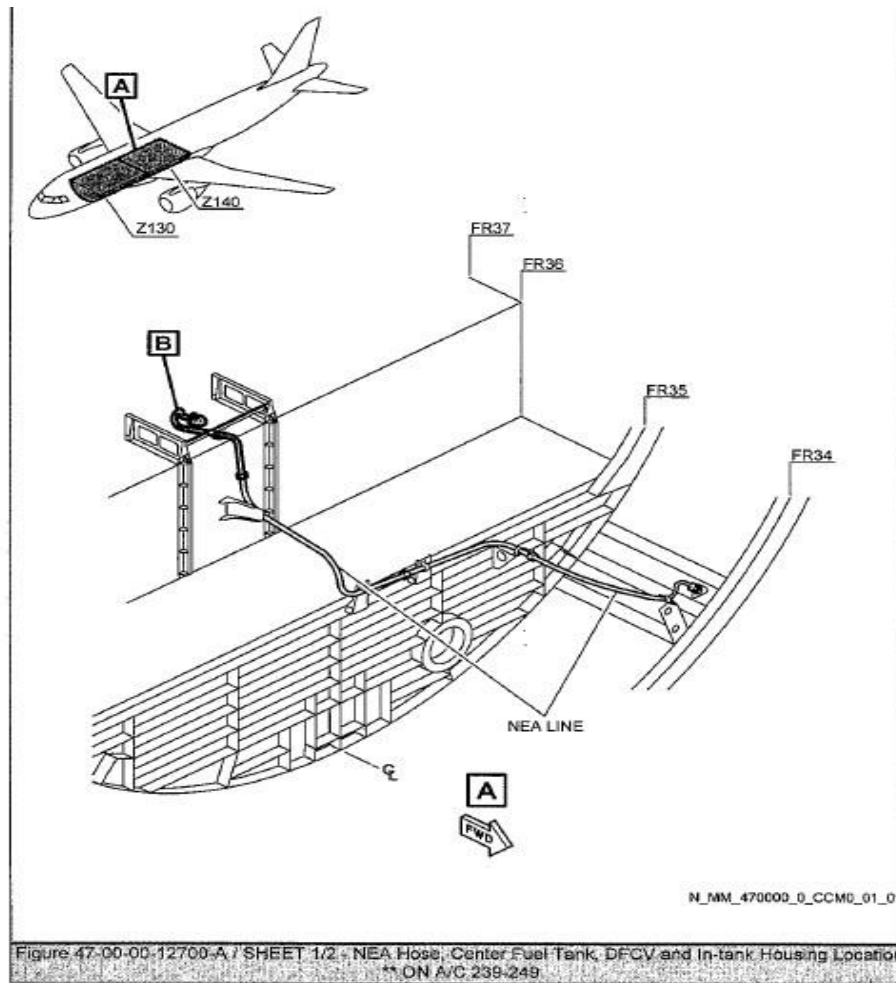


Figure 47-00-00-12700-A, SHEET 1/2 - NEA Hose, Center Fuel Tank, DFCV and In-tank Housing Location  
\*\* ON A/C 239-249

μ : 5.4 μ μ μ - :

**Airbus Aircraft Maintenance Manual**

**- Dual-Flapper Check Valve**

dual-flappercheckvalve (DFCV) μ

μ , μ μ DFCV μ μ

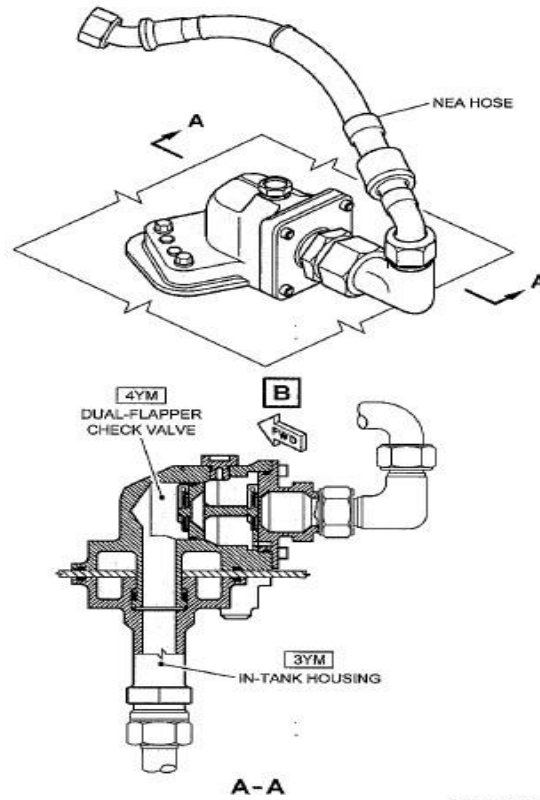
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0,29psi (0,0200 bar).

DFCV μ DFCV μ

μ checkvalve μ , μ

μ DFCV.



N\_MM\_470006\_0\_CCM0\_02\_00

Figure 47-00-00-12700-A / SHEET 2/2 - NEA Hose, Center Fuel Tank, DFCV and In-tank Housing Location  
 \*\* ON A/C 238-249

μ : 5.6 Dual Flapper Check Valve μ – : Airbus Aircraft Maintenance Manual

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μ μ 2 , μ μ μ (DC)

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μ :

1. μ 15 psi (1,0342 bar), /
2. μ μ 85 C

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7. μ FTIS

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75 C, IGGS μ μ

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• μ GGS 75 C

85 C, μ μ

• μ IGGS 85 C 60 psi

(4,1369 bar).

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90 C

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NGSSystem - NGS

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μ NGS(NitrogenGenerationSystem),

μ Boeing, μ Honeywell HamiltonSundstrand,

μ Boeing μ Airbus μ μ

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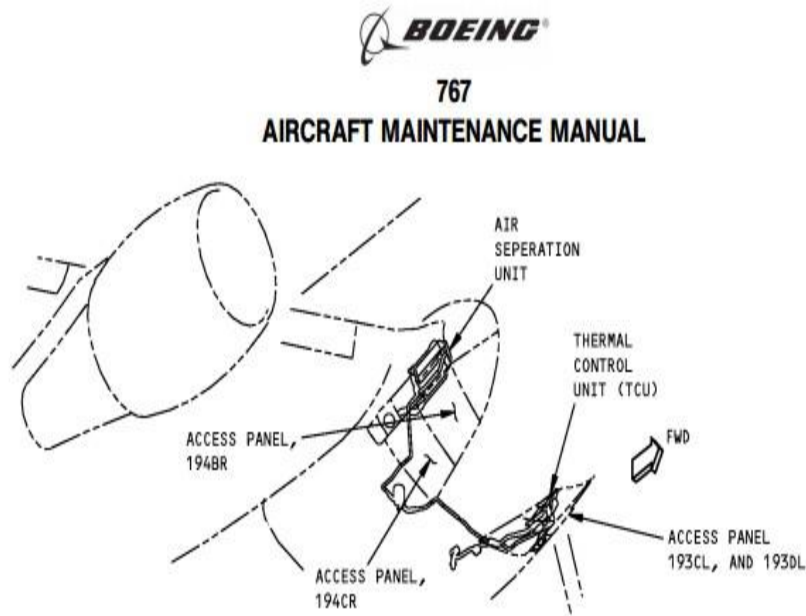
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(1) μ (TCU)

- (2) μ (AirSeparationUnit - ASU)
- (3) μ – (Nitrogen Enriched Air Distribution - NEADS)
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μ : 5.7 μ NGS 767 – : Boeing AMM

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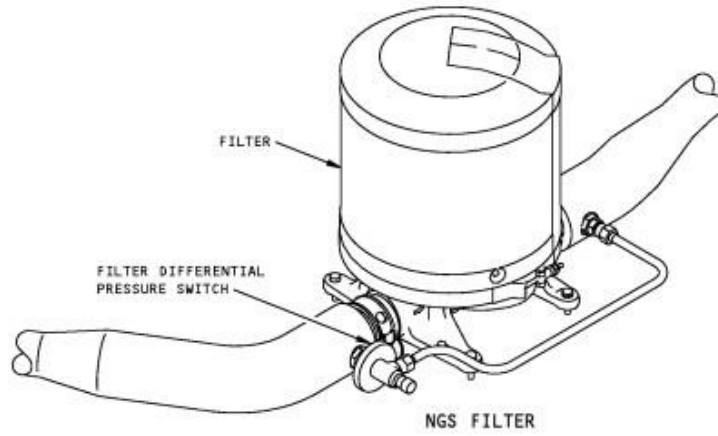


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**BOEING**  
**767**  
**AIRCRAFT MAINTENANCE MANUAL**



μ : 5.10 : Boeing AMM

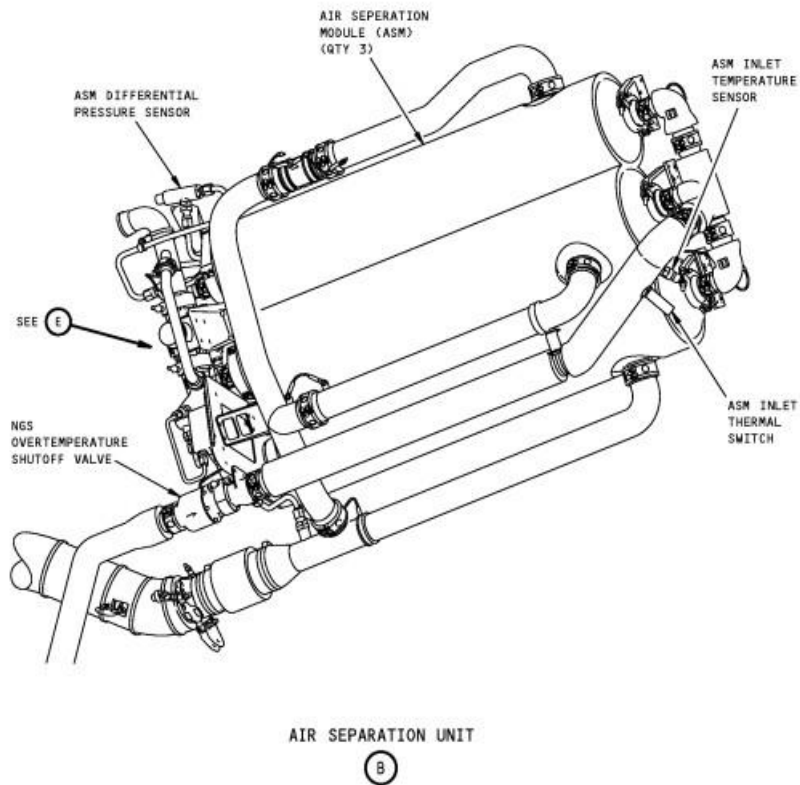
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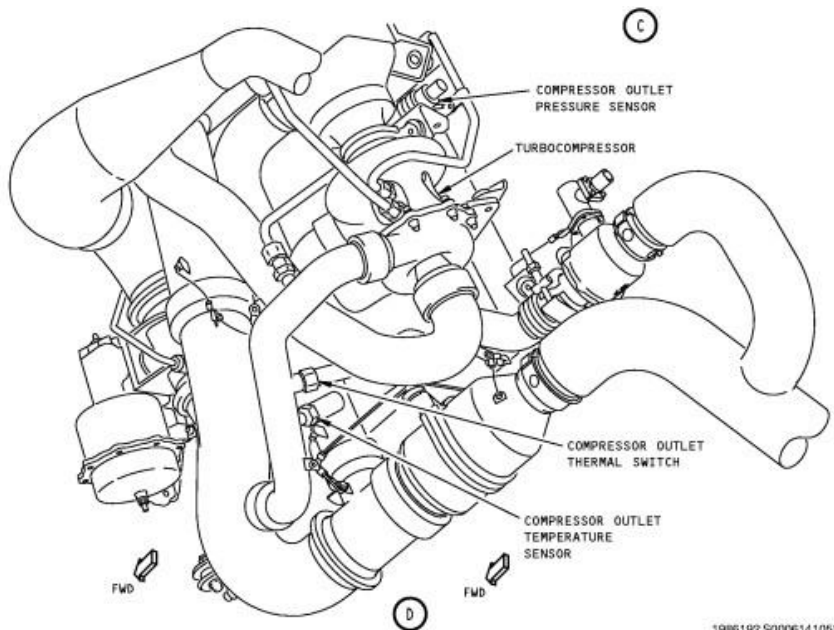
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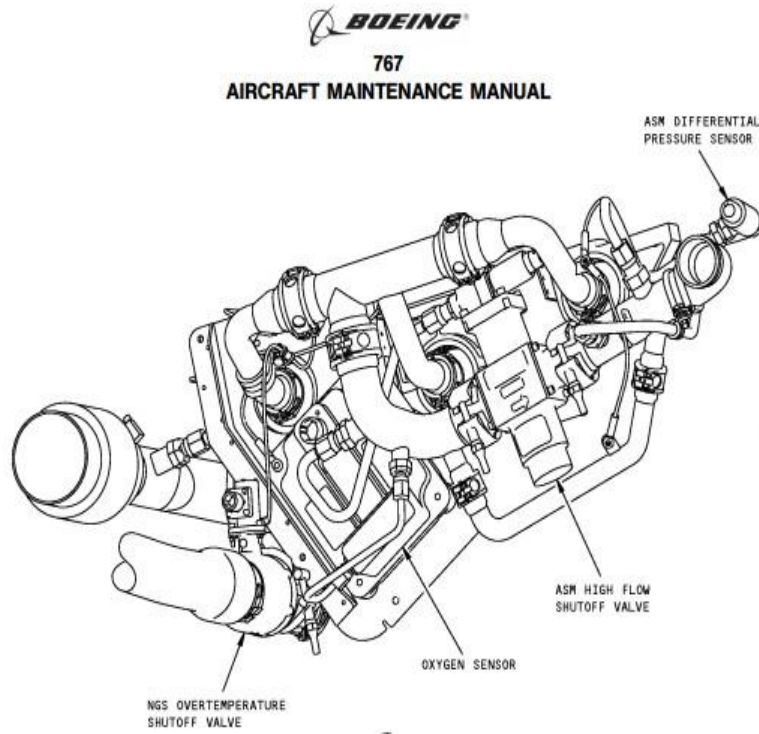
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**Inert Gas System - Description and Operation**  
**Figure 1 (Sheet 4 of 5)/47-00-00-990-801**



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: Boeing AMM









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μ :7.2 : forever wingman.com

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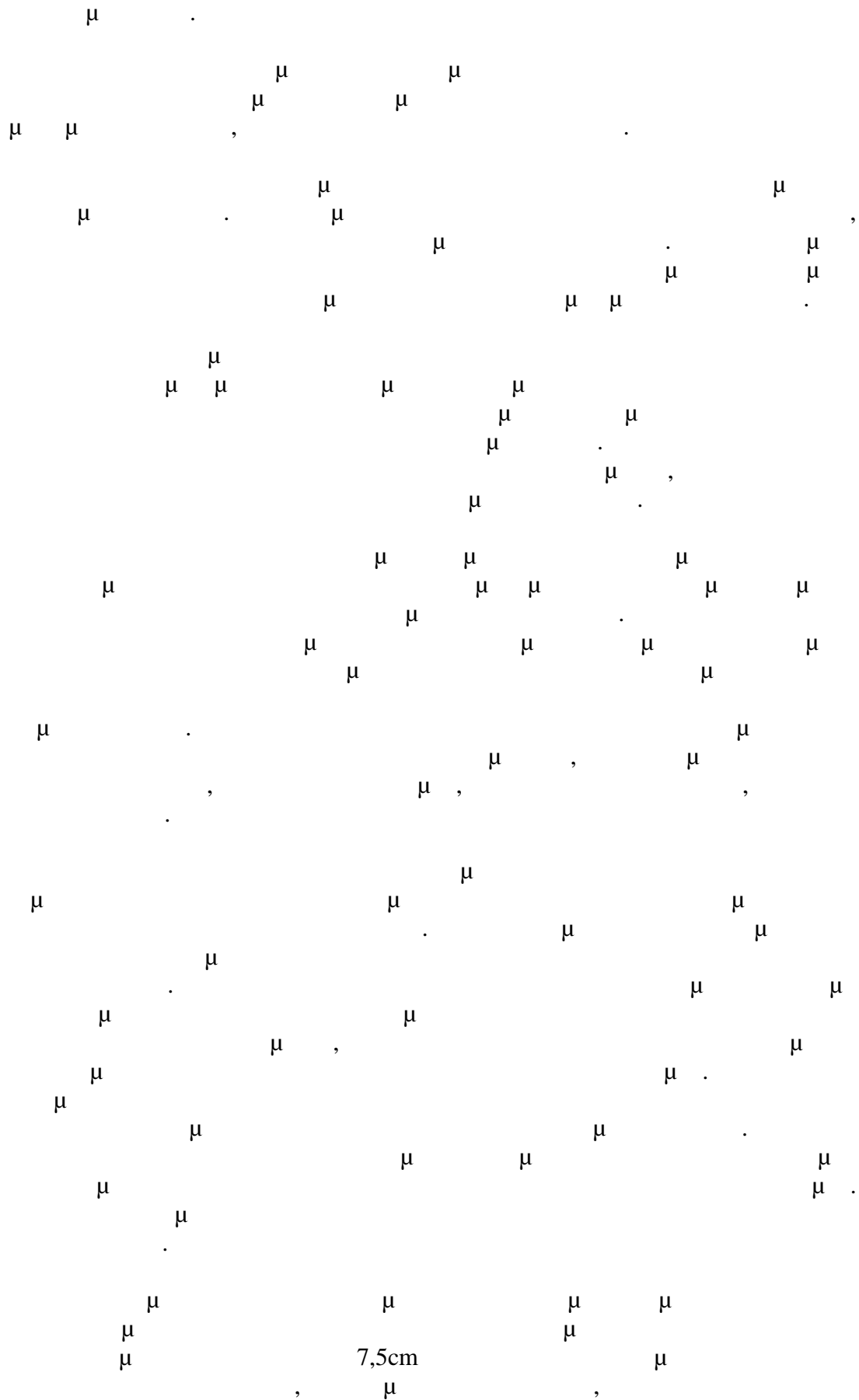








25%



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495kg,

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μ MTOW μ

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## Ασφάλεια Δεξαμενής Καυσίμων

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- . . = μ μ
- FAA = Federal Aviation Administration = μ μ
- EASA = European Aviation Safety Agency =
- NTSB = National Transportation Safety Board = μ  
μ
- TWA = Trans World Airlines = μ
- . .
- CWT = Center Wing Tank = μ μ /
- FQIS = Fuel Quantity Indicating System = μ μ μ
- AFSSP = Aircraft Fuel System Safety Program
- FTIS = Fuel Tank Inerting System = μ μ μ (Airbus)
- NGS = Nitrogen Generating System = μ (Boeing)
- ADIRU = Air Data Inertial Reference Unit = μ
- IGGS = Inert Gas Generating System = μ
- NEA = Nitrogen Enriched Air = μ μ μ
- OEA = Oxygen Enriched Air = μ μ μ
- PPE = Personal Protective Equipment = μ

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- Wikipedia
- Internet